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Lee

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- (54) **MULTI-SWIRLER FUEL/AIR MIXER WITH CENTRALIZED FUEL INJECTION**
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- (*) Notice: Subject to any disclaimer, the term of this patent is extended or adjusted under 35 U.S.C. 154(b) by 480 days.

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CPC **F23R 3/286** (2013.01); **F23R 3/14** (2013.01); **Y02T 50/675** (2013.01)
- (58) **Field of Classification Search**
CPC F23R 3/286; F23R 3/12; F23C 2900/07001; F23D 11/102; F23D 11/104; F23D 11/106; F23D 11/107; F23D 11/108
See application file for complete search history.

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(57) **ABSTRACT**

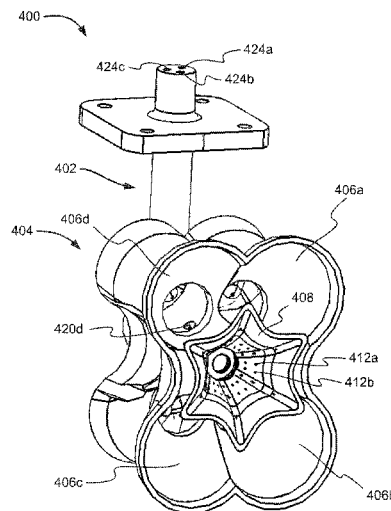
A gas turbine combustor assembly has a fuel/air mixer assembly with a plurality of fuel/air mixer elements. Each fuel/air mixer element defines an air flow passage there-through. A fuel injector is coupled to the fuel/air mixer assembly. The fuel injector has a tip portion with a plurality of fuel outlets arranged to direct fuel into the air flow passages of the air/fuel mixer elements. Each of the fuel/air mixer elements has at least one outlet arranged to supply fuel to the element.

10 Claims, 12 Drawing Sheets

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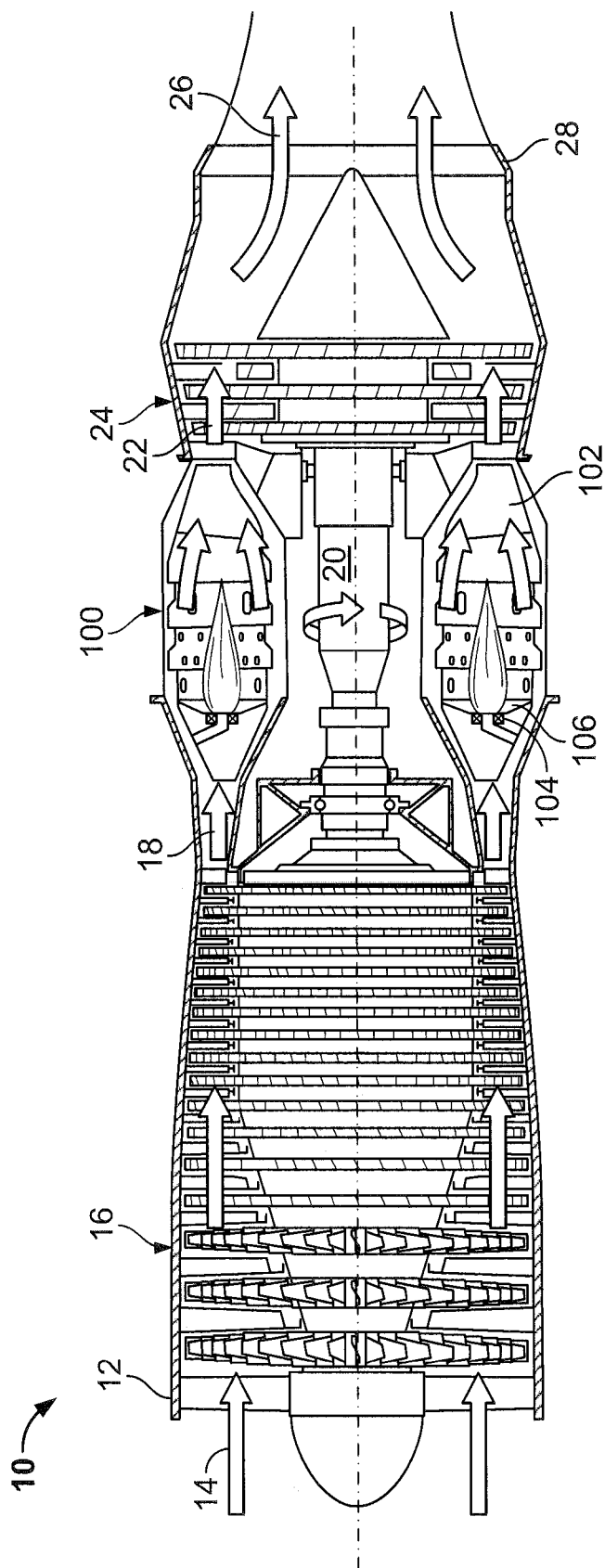


FIG. 1

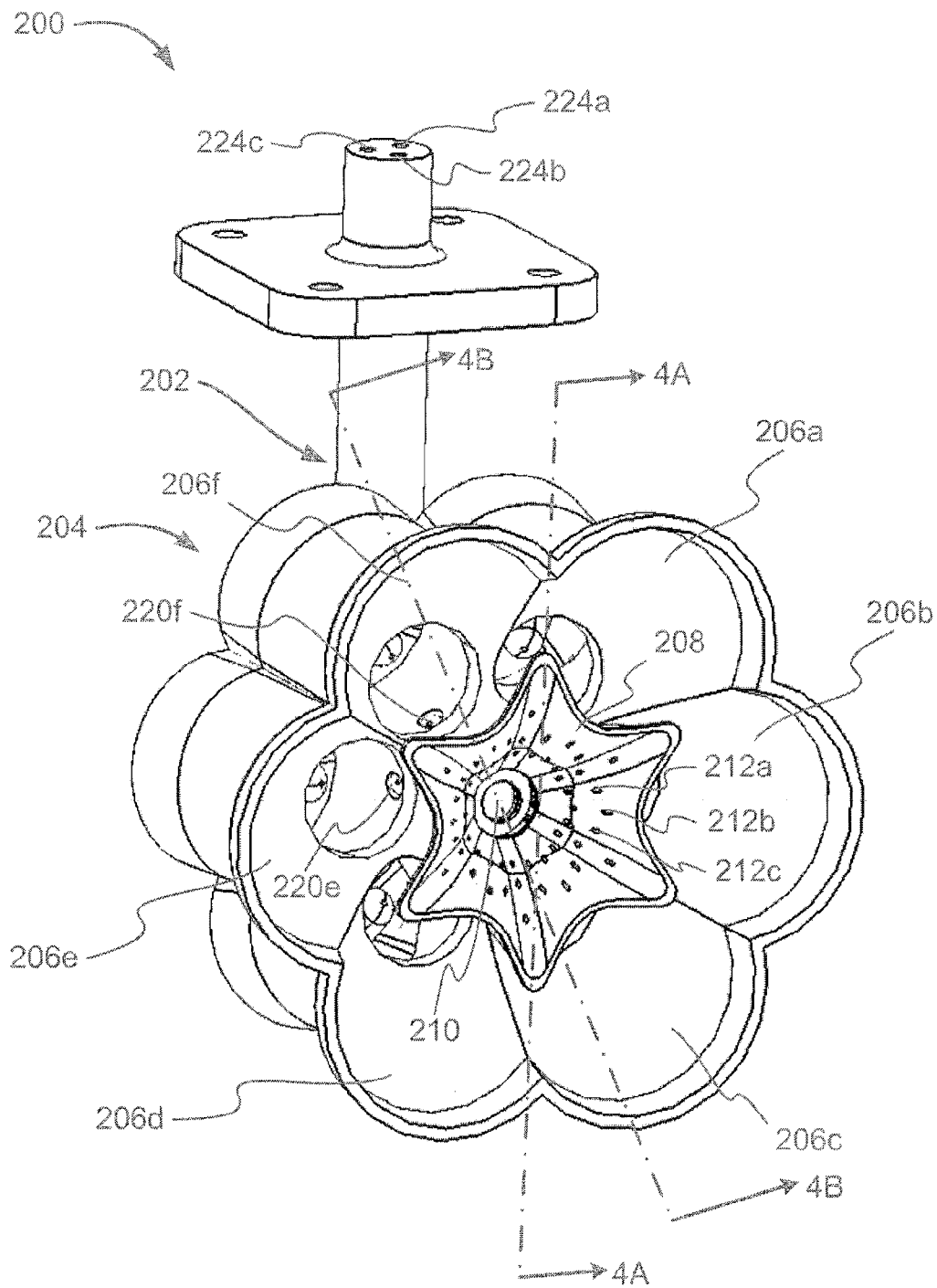


FIG. 2A

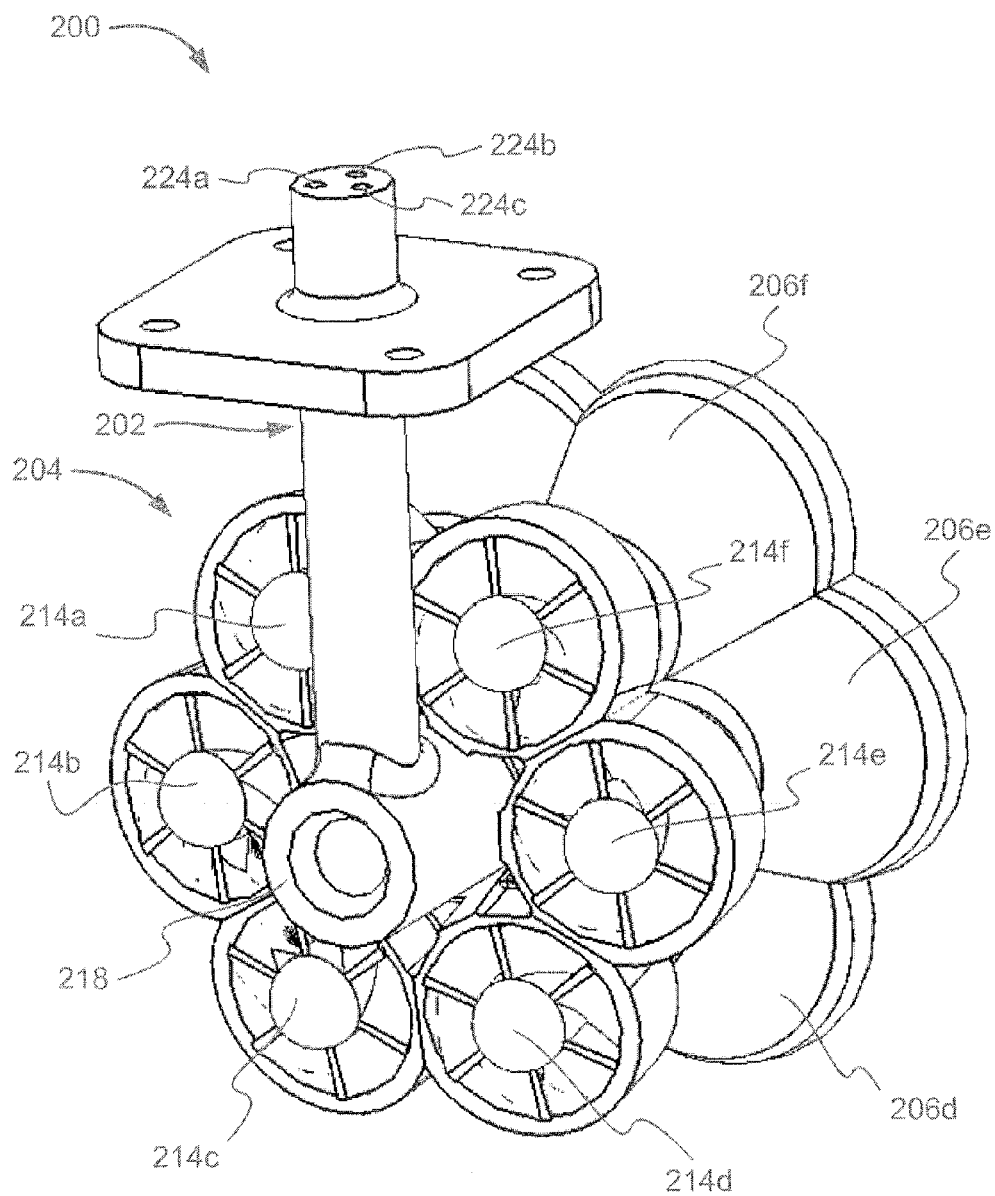


FIG. 2B

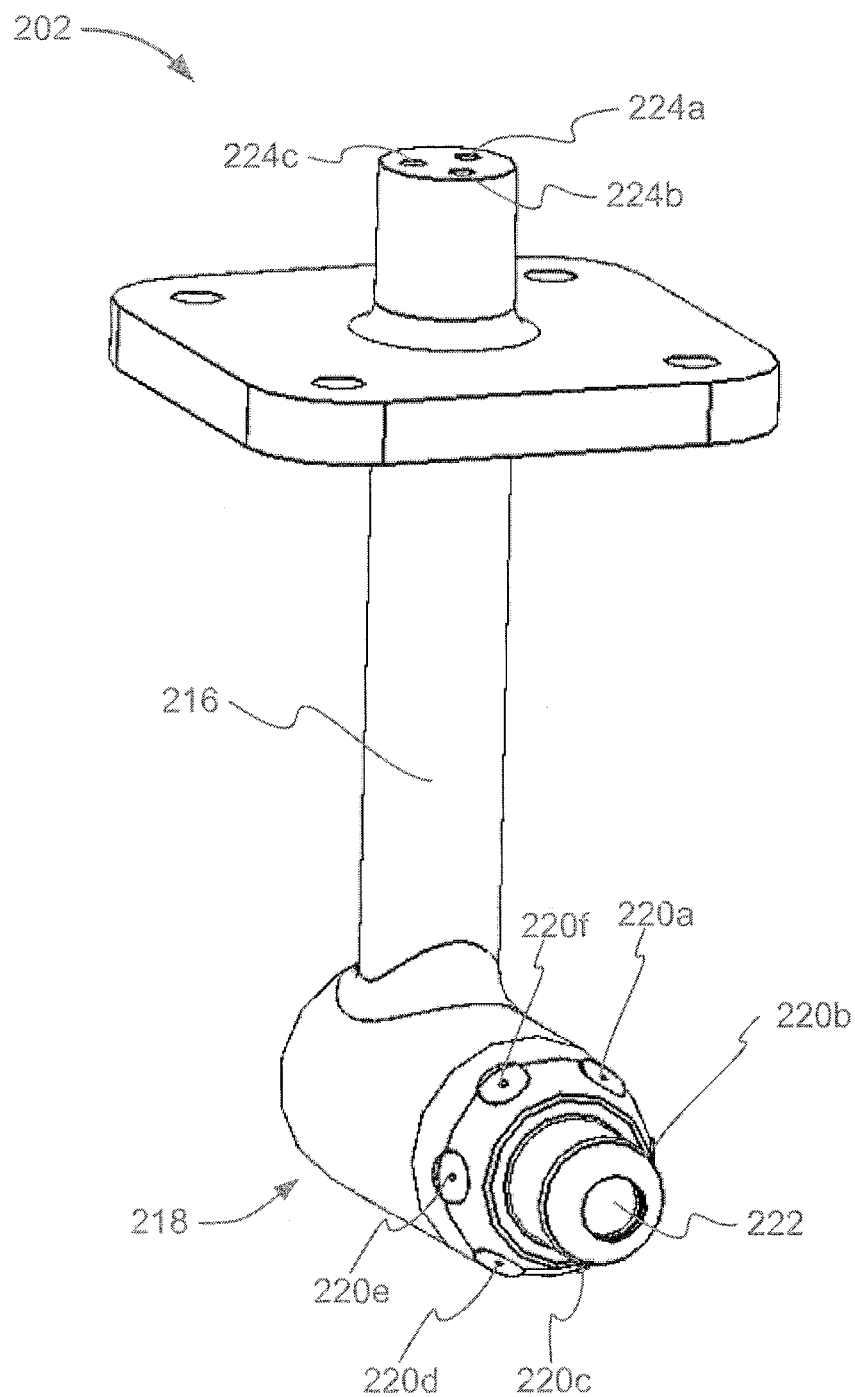


FIG. 3

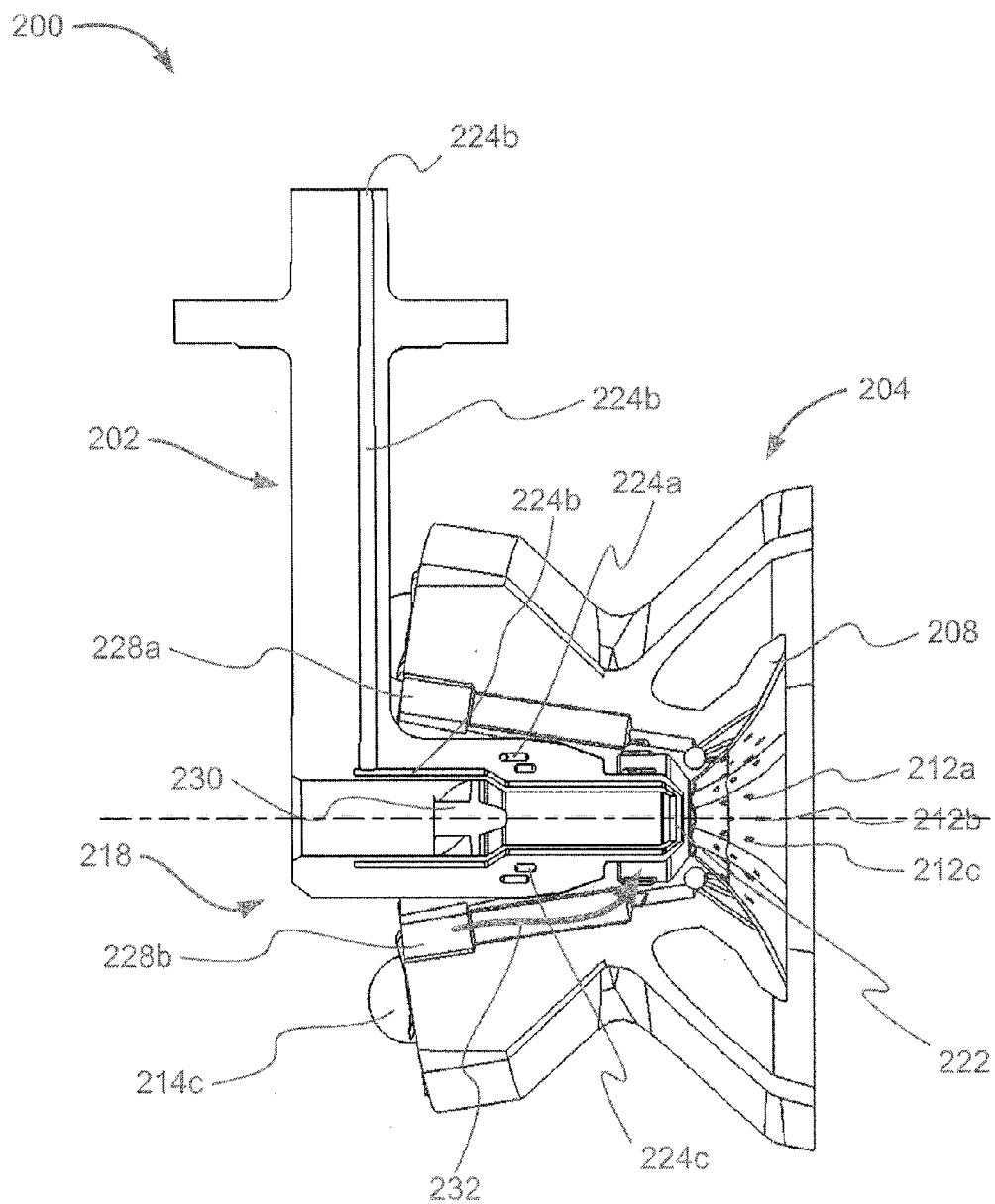


FIG. 4A

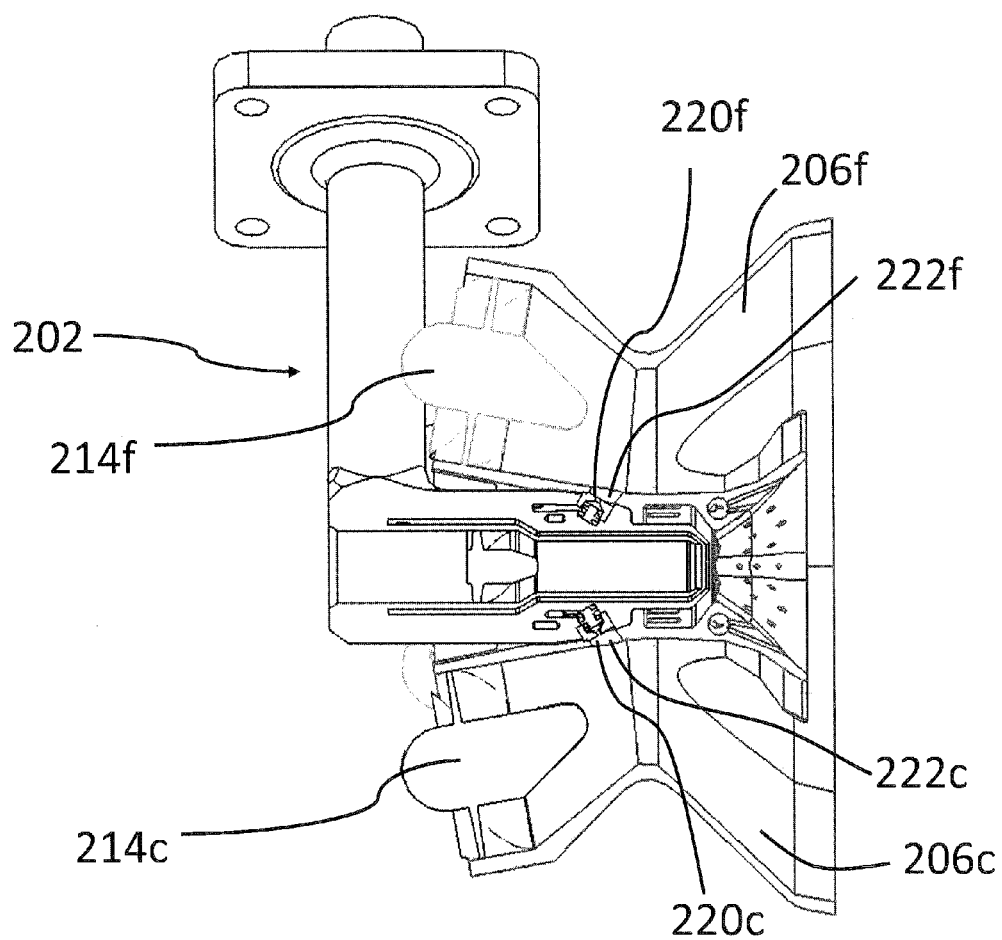


FIG. 4B

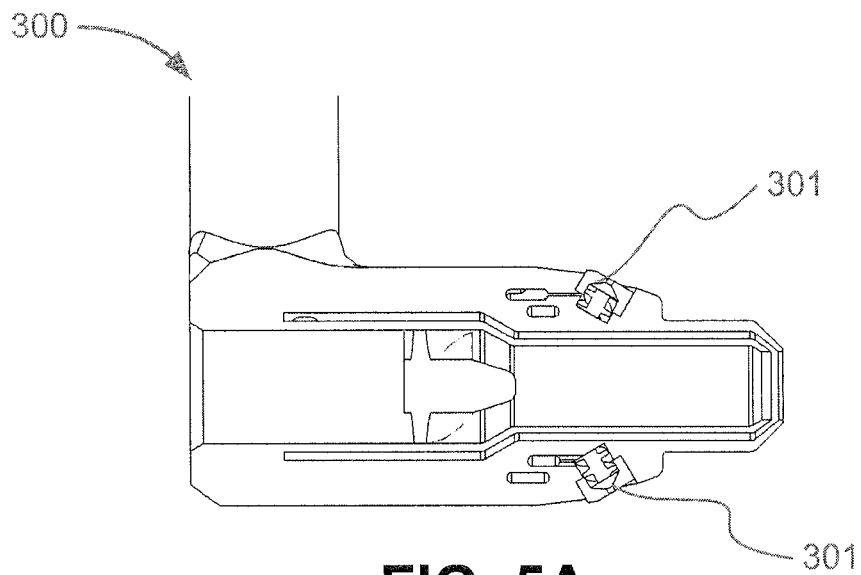


FIG. 5A

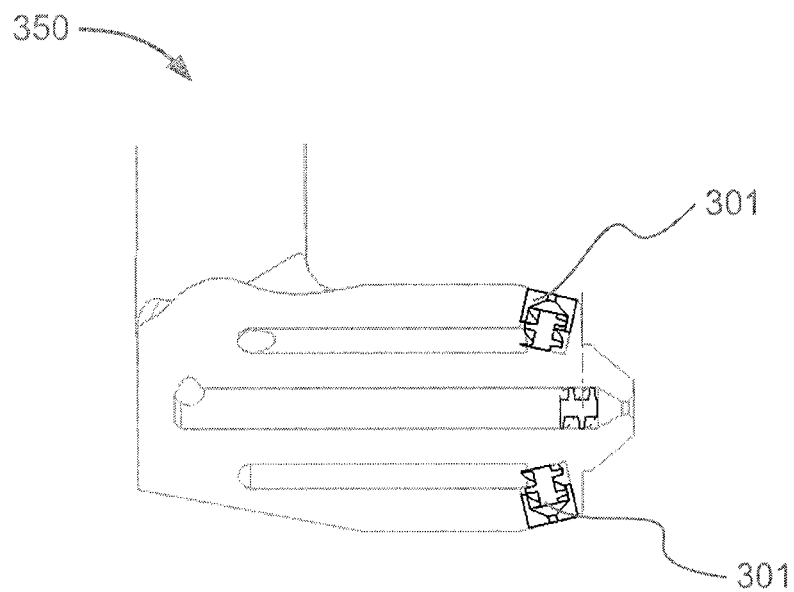


FIG. 5B

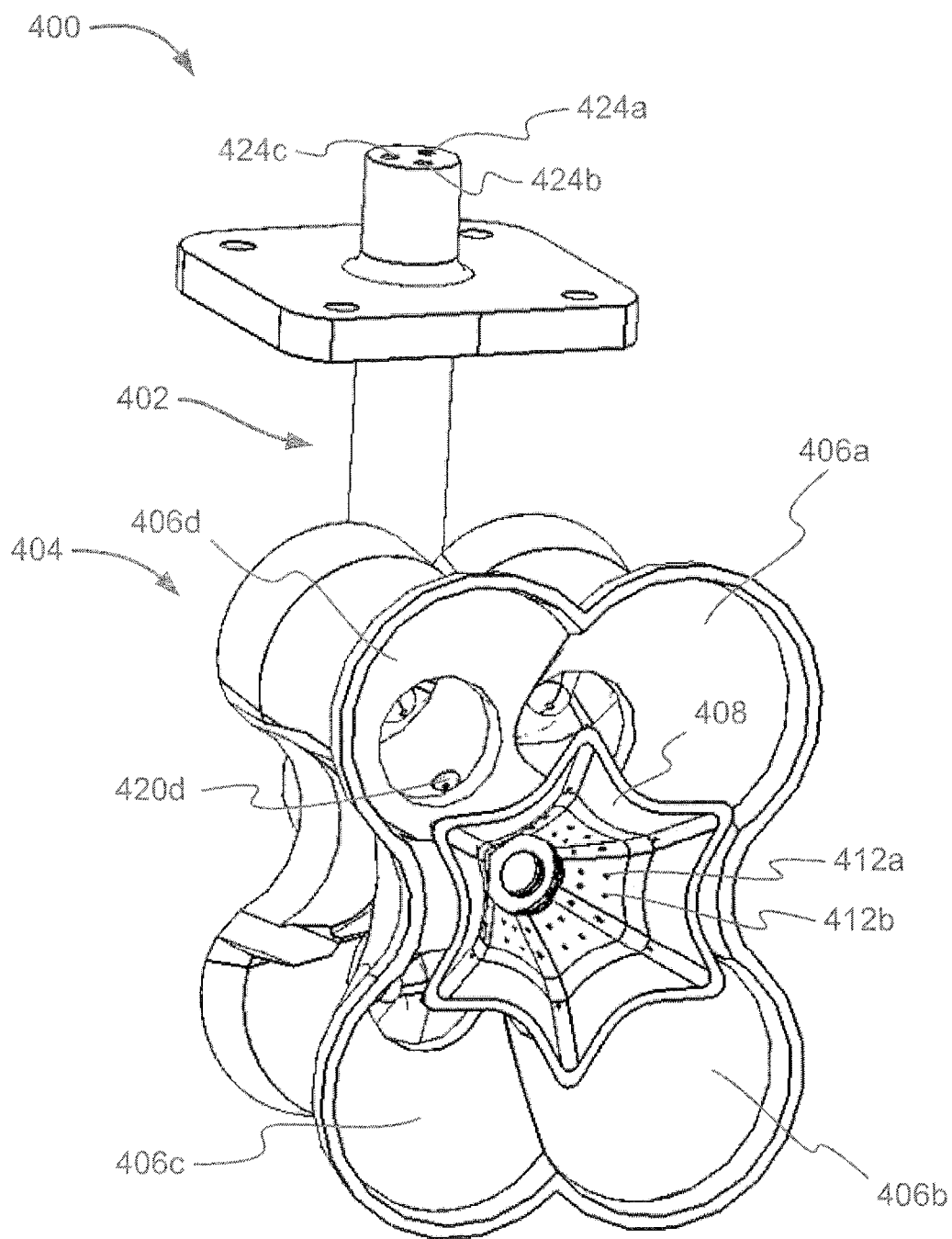


FIG. 6

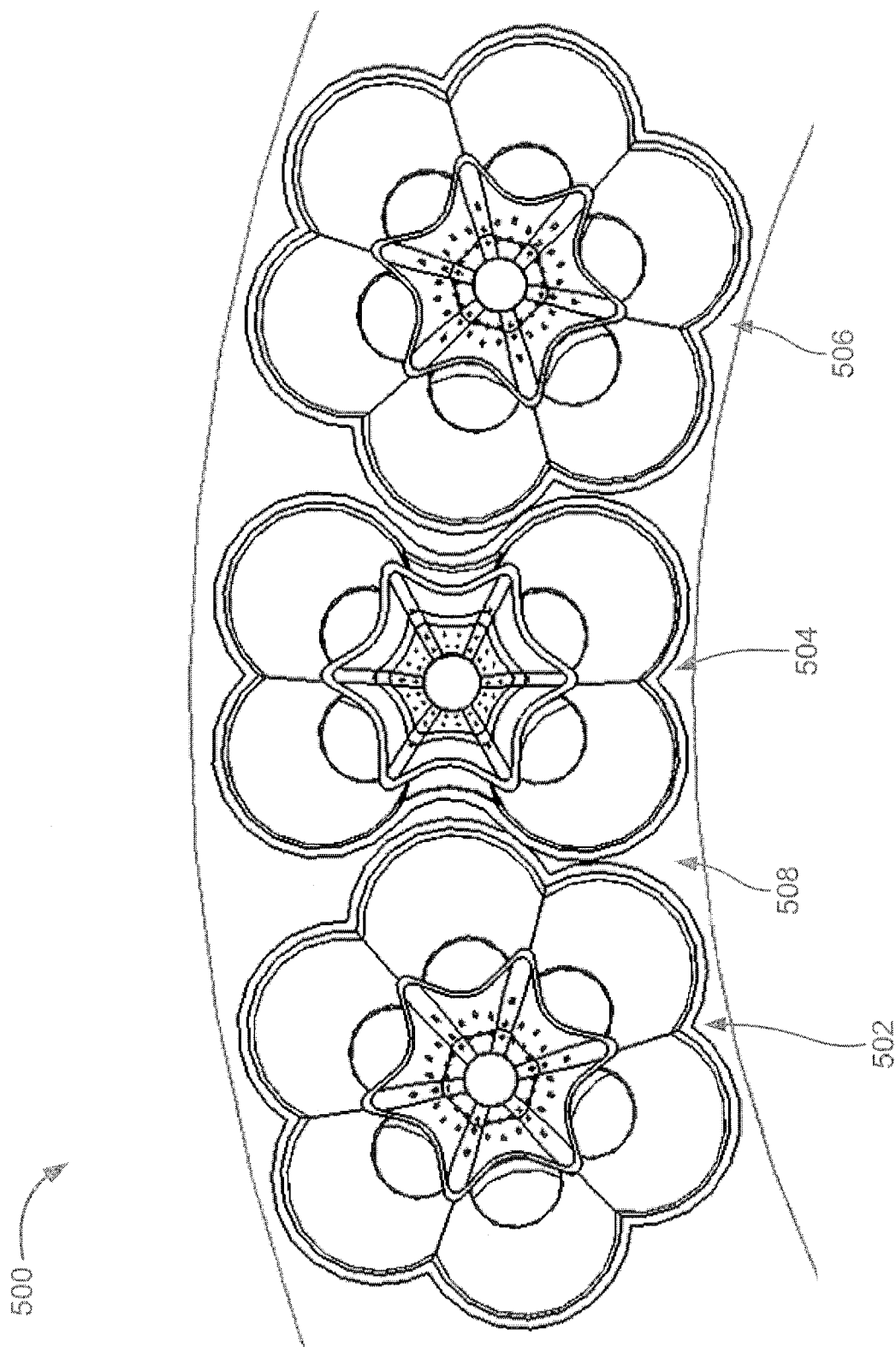


FIG. 7

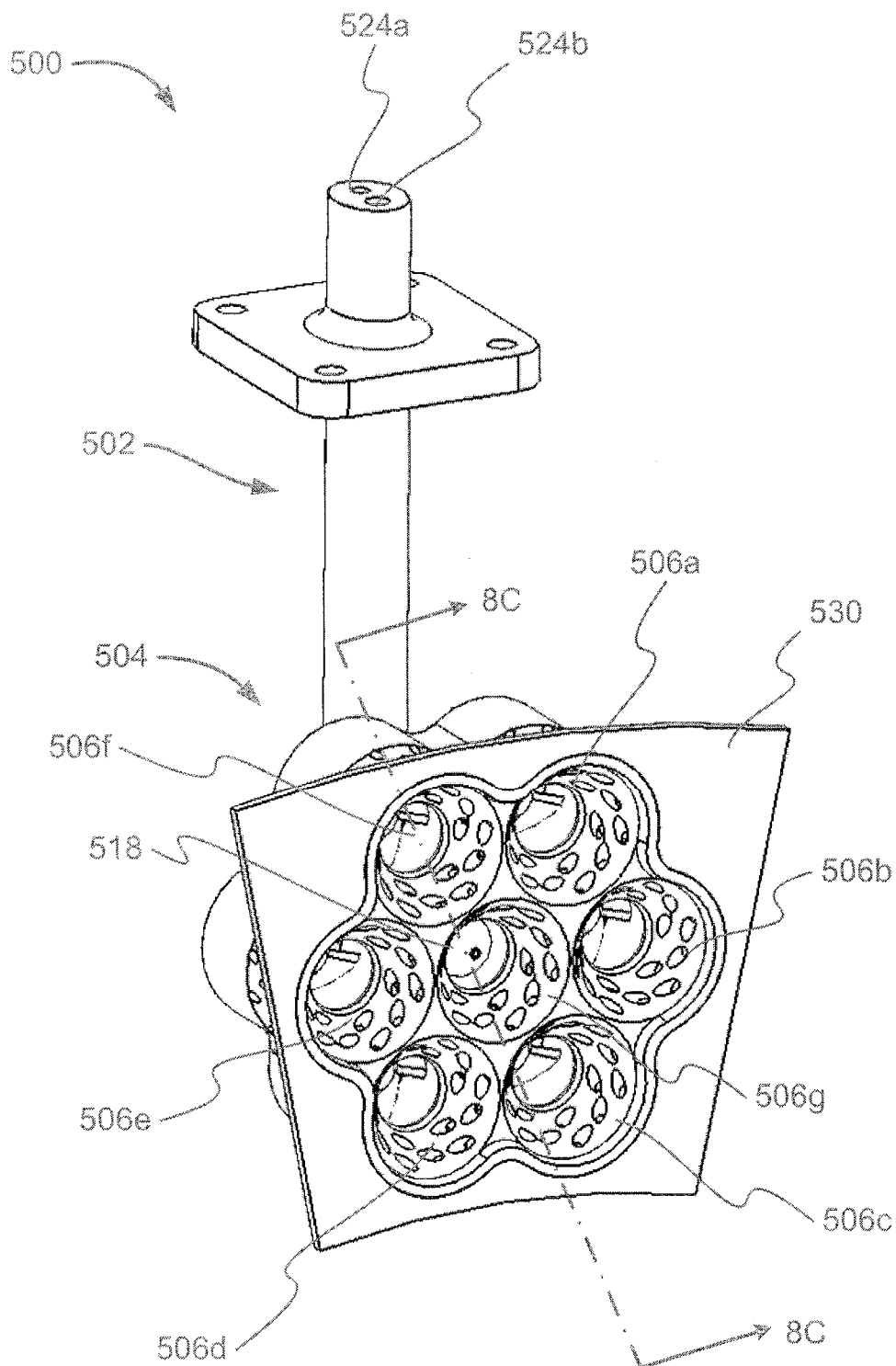


FIG. 8A

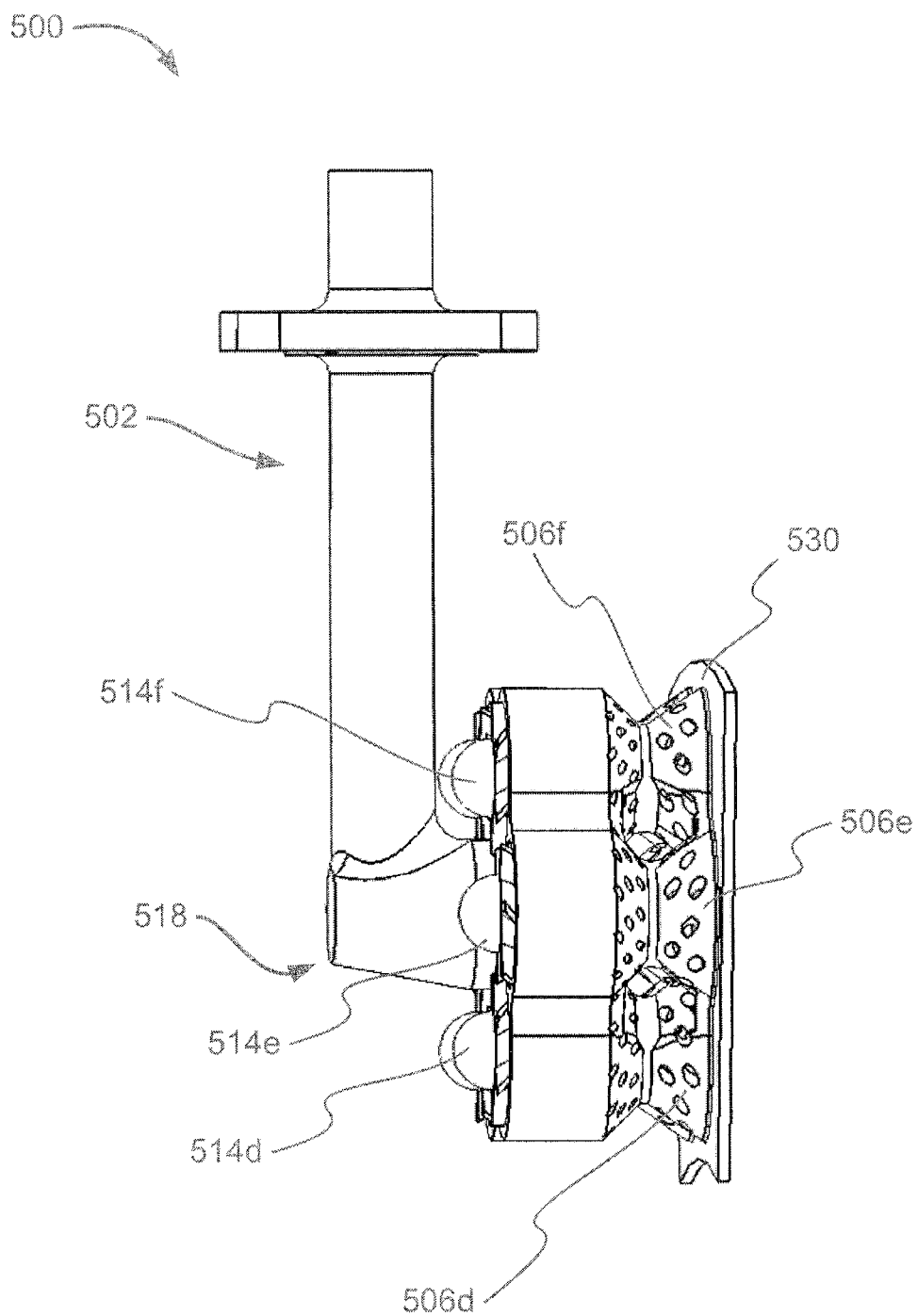


FIG. 8B

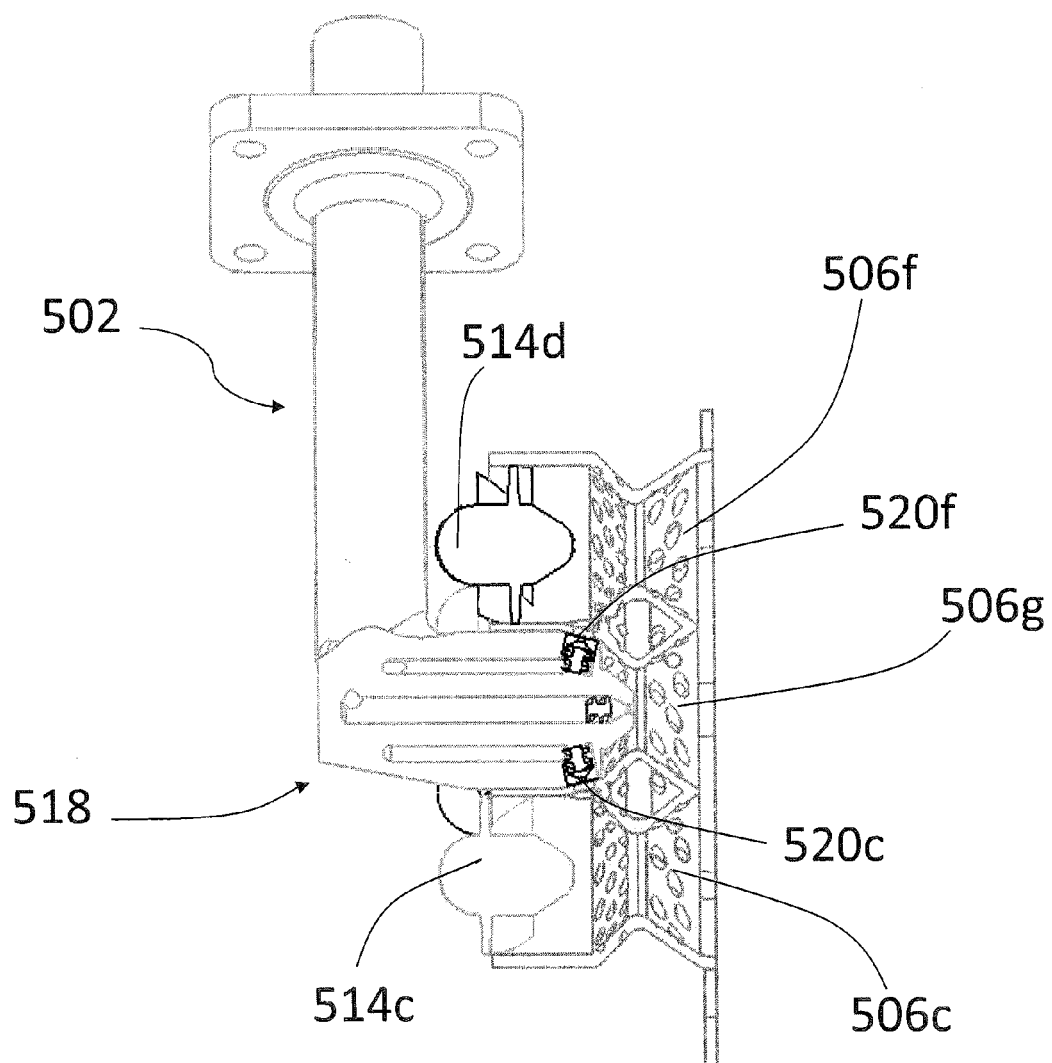


FIG. 8C

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MULTI-SWIRLER FUEL/AIR MIXER WITH CENTRALIZED FUEL INJECTION

BACKGROUND

The gas turbine engine is the preferred class of internal combustion engine for many high power applications. Fundamentally, the gas turbine engine features an upstream rotating compressor coupled to a downstream turbine, and a combustion chamber in-between. The combustion chamber can include multiple mixers to mix fuel and air prior to combustion. Typically, each mixer requires an individual fuel injector tip to spray the fuel into the mixer's airflow.

DESCRIPTION OF DRAWINGS

FIG. 1 is a half, side cross-sectional view of an example gas turbine engine.

FIG. 2A is a front perspective view of an example gas turbine combustor sub-assembly, including a fuel/air mixer assembly and fuel injector assembly.

FIG. 2B is a rear perspective view of the example gas turbine combustor sub-assembly of FIG. 2A.

FIG. 3 is a front perspective view of an example fuel injector assembly.

FIG. 4A is half, side cross-sectional view of the example sub assembly of FIGS. 2A and 2B.

FIG. 4B is half, side cross-sectional view of the example sub assembly of FIGS. 2A and 2B.

FIG. 5A is a detail half, side cross-sectional view of an example fuel injector tip.

FIG. 5B is a detail half, side cross-sectional view of an example fuel injector tip.

FIG. 6 is a front perspective view of another example gas turbine combustor sub-assembly.

FIG. 7 is a view of a portion of an annular combustor assembly.

FIG. 8A is a front perspective view of another example gas turbine combustor sub-assembly.

FIG. 8B is a side view of the example gas turbine combustor sub-assembly of FIG. 8A.

FIG. 8C is half, side cross-sectional view of the example subassembly of FIGS. 8A and 8B.

Like reference symbols in the various drawings indicate like elements.

DETAILED DESCRIPTION

FIG. 1 is a half, side cross-sectional view of an example gas turbine engine 10. The gas turbine engine 10 is turbojet-type gas turbine that could be used, for example, to power jet aircraft. However, the concepts described in the present disclosure are not so limited, and can be incorporated in the design of various other types of gas turbine engines (e.g., turbofan, turboprop, turboshaft, industrial/marine engines or others).

As shown, the gas turbine engine 10 generally facilitates a continuous axial flow of gas. That is, gas generally flows through the engine 10 in the axially downstream direction indicated by the arrows in FIG. 1. The gas turbine engine 10 includes an intake 12 that receives ambient air 14 and directs the ambient air to a compressor 16. The ambient air 14 is drawn through multiple stages of the compressor 16. High-pressure air 18 exiting the compressor 16 is introduced to a combustor 100. In certain instances the combustor 100 is an

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annular combustor circumscribing the engine's main shaft 20 or a can-type combustor positioned radially outward of the shaft.

The combustor 100 includes a combustion shield 102, multiple fuel injectors 104, and a combustor dome 106. At the combustor 100, the high-pressure air 18 is mixed with liquid hydrocarbon fuel (not shown) and ignited to produce heated combustion products 22. The combustion products 22 are passed through multiple stages of a turbine 24. The turbine 24 extracts energy from the high-pressure, high-temperature combustion products 22. Energy extracted from the combustion products 22 by the turbine 24 drives the compressor 16, which is coupled to the turbine by the main shaft 20. Exhaust gas 26 leaving the turbine 24 is accelerated into the atmosphere through an exhaust nozzle 28 to provide thrust or propulsion power.

FIG. 2A and FIG. 2B are front and rear perspective views, respectively, of an example gas turbine combustor sub-assembly 200. The example turbine combustor sub-assembly 200 mixes air and fuel and enables combustion of the air/fuel mixture within a combustor. Multiple of the sub-assemblies 200 are arranged together to form a combustor assembly. The turbine combustor sub-assembly 200 can be implemented in a gas turbine engine such as gas turbine engine 10 shown in FIG. 1. The example gas turbine combustor sub-assembly 200 includes an example fuel injector 202 and an example fuel/air mixer assembly 204. The mixer assembly 204 includes a plurality of mixer elements 206a, 206b, 206c, 206d, 206e, 206f and a center element 208. In other implementations, the mixer assembly 204 can include more or fewer mixer elements. In the example fuel/air mixer assembly 204, the mixer elements 206a, 206b, 206c, 206d, 206e, 206f are affixed to each other as well as to the center element 208 and located symmetrically around the center element 208, equidistant from a center location. The mixer elements 206a-f can be located in other configurations (e.g. asymmetrically). The mixer assembly 204 can be coupled to the fuel injector (e.g. fuel injector 104) to be carried with the fuel injector or it can be coupled to a combustor dome (e.g. dome 106) and the injector subsequently coupled to the mixer assembly 204. In some cases, the mixer elements 206a-f are coupled to the dome plate (not shown) of the dome 106.

The example mixer elements 206a-f define internal swirl passages that facilitate the mixing of air and fuel for combustion. In the example fuel/air mixer assembly 204, the example mixer elements 206a-f are conical-cylindrical-conical, converging-diverging venturis. In other implementations, the mixer elements 206a-f can be venturis with a different shape or in a different configuration. In other implementations, the mixer elements 206a-f need not be venturis and can include passages of a different shape or in a different configuration. The mixer elements 206a-f each include an air swirler 214a-f, respectively. The air swirlers 214a-f are located within the passages and impart a swirling motion on the incoming air that facilitates the mixing of air and fuel. In the example mixer assembly 204 the air swirlers are axial fans, though other configurations or types of air swirlers can be used.

The center element 208 defines a central mixer passage 210 that enables mixing of airflow and fuel. Fuel is fed from the fuel injector assembly 202 into the central mixer passage 210 and the fuel/air mixture from the central mixer passage 210 is combusted. In some implementations, the back surface of the center element 208 defines a flame holder. The fuel/air mixture output from the mixer elements 206a-f can be ignited by a standing flame held in place by the back

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surface of center element **208**. The center element **208** includes a plurality of cooling air passages. (e.g. cooling air passages **212a**, **212b**, **212c**, among others). The airflow through the cooling passages helps maintain the center element **208** within acceptable temperatures for the materials of the element. Furthermore, the airflow from the cooling air passages mixes with the fuel output from the central mixer passage **210**.

FIG. 3 shows a front perspective view of the example fuel injector assembly **202** of FIG. 2A and FIG. 2B. The fuel injector assembly **202** includes a stem **216** with a securing flange and an example tip portion **218**. The tip portion **218** includes the extreme tip of the fuel injector. The tip portion **218** includes multiple peripheral fuel orifices **220a-f** and a central orifice **222**. The tip portion **218** can be a single tube having a cylindrical shape or some other shape. The tip portion **218** includes one or more internal passages that feed fuel to the orifices **220a-f**, **222**.

The peripheral orifices **220a-f** and central orifice **222** spray or inject liquid or gaseous fuel, and/or other fluids such as water or steam into the airflow of mixer assembly **204**. The orifices **220a-f** and **222** can be designed to produce a specified spray pattern, for example to produce a jet, a hollow or solid cone, or another specific shape (e.g., fan or other shape). The peripheral orifices **220a-f** are located around the tip portion **218**, oriented radially outward, such that each peripheral orifice **220a-f** is positioned in the sidewall of a mixer element **206a-f** (c.f. FIG. 2A). The peripheral orifices **220a-f** are configured to spray fuel into the airflow of associated mixer elements **206a-f**, respectively. The central orifice **222** sprays fuel into the central mixer passage **210** and thus into the flow of center element **208**. The plurality of orifices **220a-f** and **222** enable the fuel injector assembly **202** to supply fuel to multiple mixer elements using only a single tip or tip portion. The orifices **220a-f** and **222** can be orifice devices that thread or are otherwise installed into the tip portion **218**, or the orifices can be integral in the sidewall of the tip portion **218** (e.g., machined, laser cut, or other).

The example fuel injector assembly **202** includes fuel feeds **224a-c** that are internal passages or tubes within the stem **216** and the tip portion **218**. The fuel feeds **224a-c** are fluidly connected to the peripheral orifices **220a-f** and central orifice **222**. The fuel feeds **224a-c** supply fuel to the orifices **220a-f** and **222**. In some implementations, the fuel feeds **224a-c** can include annular passages within the tip portion **218**. The fuel feeds **224a-c** and the orifices **220a-f** and **222** can be connected such that one fuel feed (e.g. **224a**) supplies fuel to a subset or group of the orifices (e.g. **220a**, **220c**, **220e**). Likewise, a separate fuel feed (e.g. **224b**) can supply fuel to another subset or group of the orifices (e.g. **220b**, **220d**, **220f**). The central orifice **222** can also have its own fuel feed (e.g. **224b**). The fuel feeds and orifices can be connected in this configuration or another configuration, and more or fewer fuel feeds or orifices can be used. In this manner, the example fuel injector assembly can selectively supply or not supply fuel to a first group of orifices (and thus a first group of mixer elements) and to a second group of orifices (and thus a second group of mixer elements) independently, thus enabling staging the mixer elements to produce different combustion conditions. The fuel injector assembly can be designed to separately control any number of groups of orifices. The fuel injector assembly can thus be a staged system designed to supply fuel independently to different groups of orifices and mixer elements.

FIG. 4A is a cross-section view of the example gas turbine combustor sub-assembly **200**. The cross-section plane

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4A-4A for FIG. 4A is shown in FIG. 2A. The cross-sectional view of FIG. 4A shows that the mixer assembly **204** includes example cooling air intake passages **228a**, **228b**. The air intake passages **228a**, **228b** are passages that allow airflow to enter the cooling air passages (e.g. **212a**, **212b**, **212c**) in the center mixer element **208**, and also provide the outer swirling air **232** for the air blast injector tip **222**. The mixer assembly **204** can include more or fewer cooling air intake passages, and can be configured differently. FIG. 4 also shows an example air swirler **230** in the tip portion **218**. The air swirler **230** is an axial fan that facilitates the inner swirling air to the airblast injector tip in the center element **208**. In other configurations, a different type of swirler could be used. FIG. 4B is a cross-section view of 4B-4B shown in FIG. 2A. The peripheral pressure atomizing tips **220c** and **220f** are each aligned with an open port **222c**, **222f** in the sidewall of the peripheral mixer elements **206c**, **206f** to facilitate the fuel injection into the airflow streams in the mixer elements.

FIG. 5A and FIG. 5B show cross-sections of two example types of fuel injector tips **300** and **350**. The example fuel injectors **300** and **350** can be implemented in a fuel injector tip portion such as the tip portion **218** shown in FIGS. 2-4. FIG. 5A shows an air-blast type fuel injector tip **300** and the peripheral pressure atomizing injection tips **301**. The example tip portion **218** of prior figures is shown including an air-blast type fuel injector tip like fuel injector tip **300**. FIG. 2B shows a pressure atomizing type fuel injector tip **350**. These and other types or combinations of fuel injector tips can be implemented in the tip portion described herein.

FIG. 6 shows an example gas turbine combustor sub-assembly **400**. The example gas turbine combustor sub-assembly **400** is substantially similar to the example gas turbine combustor sub-assembly **200** show previously. The example gas turbine combustor sub-assembly **400** includes an example fuel injector **402** and an example fuel/air mixer assembly **404**. The mixer assembly **404** includes a plurality of mixer elements **406a**, **406b**, **406c**, **406d**, and a center element **408**. The gas turbine combustor assembly **400** includes air cooling passages **412a**, **412b** and fuel or fluid feeds **424a-c**. The gas fuel injector **402** includes multiple orifices (e.g. orifice **420d**) that are configured to provide fuel to the mixer elements **406a-d** and **408**. The mixer assembly **404** includes five mixer elements (four symmetrically around and equidistant from a center element), whereas the previous mixer assembly **204** includes seven mixer elements. As discussed below, the mixer assembly **204** and mixer assembly **404** are arranged to nest together.

FIG. 7 shows a portion of an annular fuel injector assembly **500**. The annular fuel injector assembly **500** is a collection of combustor sub-assemblies that have been positioned in an annular arrangement. In certain instances, the assembly **500** can include multiple concentric annular arrangements of combustor sub-assemblies. The portion of the annular fuel injector assembly **500** depicted shows three combustor sub-assemblies **502**, **504**, **506** behind a dome plate assembly **508**. Mixer assemblies **502** and **506** are substantially similar to previously shown mixer assembly **204**, and mixer assembly **504** is substantially similar to previously shown mixer assembly **404**. The mixer elements, such as mixer elements **406a-d**, are arranged in mixer assembly **504** such that a space is available to accommodate a protruding mixer element, such as the elements **206b** and **206e**, on mixer assemblies **502** and **506**. As such, a combination of mixer assemblies **204** and mixer assemblies **404** can be tiled or nested to fill a space, as shown in FIG. 7. In some implementations, multiple arrangements of annular fuel injector

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assemblies can be used. In some implementations, the mixer elements can be arranged to fill a space other than an annulus (for example, a circular space). The mixer assemblies can also have different arrangements of mixer elements such that the mixer assemblies nest differently than shown.

FIG. 8A, FIG. 8B, and FIG. 8C show another example gas turbine combustor sub-assembly 500. The combustor sub-assembly 500 is similar to the example gas turbine combustor sub-assembly 200 shown in FIG. 2-4. The combustor sub-assembly 500 includes a fuel injector assembly 502 and a mixer assembly 504. The fuel injector assembly 502 includes multiple fuel feeds 524a and 524b and an injector tip portion 518. The mixer assembly 504 includes mixer elements 506a-g. As with combustor sub-assembly 200, a single fuel injector 202 with a single tip portion 518 is used to supply fuel to multiple mixer elements. The example mixer assembly 500 is coupled to a dome plate 530. The dome plate 530 is a section of the dome of a combustor, such as dome 106 shown in FIG. 1. The injector tip portion 518 can contain multiple orifices to direct fuel into the airflow of each mixer element 506a-g. In this example, the fuel injector tip portion 518 uses a pressure-atomizing design, though other fuel injector tip types can be used. The mixer assembly can have more or fewer mixer elements, and the mixer elements may be configured or arranged so as to nest next to other mixer elements or mixer assemblies.

The mixer elements 506a-g define venturis or passages that direct airflow into the combustor and promote fuel/air mixing. Each mixer element 506a-g includes a plurality of angled air holes in its walls. The angled air holes provide mixer element wall cooling and additional swirling airflow for fuel/air mixing. The angled air holes can have a symmetrical arrangement as shown or have another arrangement. The angled air holes can also be located at different locations on the passages of the mixer elements. The mixer elements 506a-g can also include air swirlers 514a-g (e.g. axial fans) located within the passages to further promote fuel/air mixing.

Notably, the concepts herein have been discussed in connection with a combustor and sub-assemblies for a gas turbine engine. However, they are not so limited. For example, the same concepts could be applied to other forms of engines, as well as other apparatus using combustors and/or combustion chambers.

While this specification contains many details, these should not be construed as limitations on the scope of what may be claimed, but rather as descriptions of features specific to particular examples. Certain features that are described in this specification in the context of separate implementations can also be combined. Conversely, various features that are described in the context of a single implementation can also be implemented in multiple embodiments separately or in any suitable subcombination.

A number of examples have been described. Nevertheless, it will be understood that various modifications can be made. Accordingly, other implementations are within the scope of the following claims.

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The invention claimed is:

1. A gas turbine combustor assembly, comprising:
 - a fuel/air mixer assembly comprising a plurality of fuel/air mixer elements, each fuel/air mixer element defining an air flow passage therethrough, the plurality of fuel/air mixer elements comprising a center fuel/air mixer element and multiple peripheral fuel/air mixer elements arranged around the center fuel/air mixer element; and
 - a fuel injector coupled to the fuel/air mixer assembly comprising a tip portion with a plurality of fuel outlets arranged to direct fuel into the air flow passages of the fuel/air mixer elements, the plurality of fuel outlets comprising a center fuel outlet and multiple peripheral fuel outlets, the center fuel outlet arranged to direct fuel into the center fuel/air mixer element and the peripheral fuel outlets arranged to direct fuel into the peripheral fuel/air mixer elements, each of the peripheral fuel/air mixer elements comprising a Venturi housing with an opening in a sidewall of the Venturi housing, and where each of the peripheral fuel outlets of the fuel injector is arranged to direct fuel through the opening in the sidewall of a respective peripheral fuel/air mixer element.
2. The gas turbine combustor assembly of claim 1, where the fuel outlets are in a sidewall of the tip portion.
3. The gas turbine combustor assembly of claim 1, where the peripheral fuel/air mixer elements are arranged around the center fuel/air mixer element, and where the fuel injector resides proximate the center fuel/air mixer element.
4. The gas turbine combustor assembly of claim 3, where the fuel injector resides in the center fuel/air mixer element.
5. The gas turbine combustor assembly of claim 3, where a center of each peripheral fuel/air mixer element is equidistant from the center fuel/air mixer element.
6. The gas turbine combustor assembly of claim 3, where the peripheral fuel/air mixer elements are symmetrically arranged about a first axis passing through the center fuel/air mixer element.
7. The gas turbine combustor assembly of claim 6, where the fuel/air mixer elements are symmetrically arranged about a second, different axis passing through the center location.
8. The gas turbine combustor assembly of claim 1, wherein the fuel/air mixer assembly comprises a first fuel/air mixer assembly, and wherein the gas turbine combustor assembly further comprises a second fuel/air mixer assembly nested with the first fuel/air mixer assembly.
9. The gas turbine combustor assembly of claim 1, where the fuel injector comprises a first fuel inlet and a second fuel inlet,
 - where a first subset of the fuel outlets is fluidically coupled to receive fuel from the first fuel inlet, and
 - where a second subset of the fuel outlets is fluidically coupled to receive fuel from the second fuel inlet.
10. The gas turbine combustor assembly of claim 1, where all fuel outlets of the fuel injector are incorporated within the tip portion of the fuel injector.

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